Airfilm Camera Systems
6245 Aerodrome Way, Hanger 2
Georgetown, CA

Document: AFNW-006
STC Number: SR02288LA

INSTALLATION INSTRUCTIONS

NOSE WINDOW UTILITY MOUNT

FOR THE
EUROCOPTER
AS 350 and AS355
SERIES HELICOPTERS

RECORD OF REVISIONS

Rev.	Page	Date	Description	FAA Approval
N/C	1-17	5/06/2010	Initial Release	FAA APPROVED MAY 6 2010 LOS ANGELES AIRCRAFT CERTIFICATION OFFICE INITIALS: .....

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1. List of Approved Payloads

The following sensor / camera / light have been installed and flown.

- GYRON MODEL 945 SERIES
- GYRON MODEL 935 SERIES

This STC addresses the Structural, Performance & Handling Qualities requirement for the largest configurations 200 lbs (90 Kg), and 3.6 square feet (3345 square cm).

The specific sensor/cameras/light not listed here is accepted with the follow-on test plan found in Appendix A.

1.1. For helicopters registered in United States or other countries recognizing FAA Certification:

Sensor / camera / lights or payloads listed above do not require further FAA flight testing.

Once the installation for a sensor/camera/payload not on the list above is completed by the Integrator/Operator and the flight test conducted by the Pilot/Operator and the FAA (certified) mechanic the sensor /camera / light payload can be added to the accepted list in this manual. The report contained herein must be completed and signed prior to the “return to service” for any sensor/ camera / light payload.

The flight will be conducted as an “Operational Check Flight”. Operational check flights do not require a special airworthiness certificate in the experimental category. The term “operational check flight” (14 CFR § 91.407(b)) includes flight tests performed to check installation and/or operation of an approved STC, amended TC, or any other FAA-approved data after installation and return to service.

Operational check flights are performed under the current airworthiness certificate.

The purpose of this test is to ensure the approved modification and/or alteration functions properly and does not adversely affect aircraft operation.

1.2. For helicopters registered in an EU-Member State:

For a specific sensor/camera/light or downlink antenna to be added to the STC, a Minor Change is required with an EASA accepted certification program.

Once the testing is completed by the Integrator/Operator and the flight test conducted by the Pilot/Operator and EASA Engineer and the Minor Change is approved the sensor/camera/light, can be added to the accepted list in this manual. The report contained herein must be completed and signed prior to the "return to service" for sensor/camera/light.

The flights have to be conducted with a "Permit to Fly".

The purpose of this test is to ensure the approved modification and/or alteration functions properly and does not adversely affect aircraft operation.

1.3. For all helicopters:

The installation is assumed to have a self-contained power supply or connected to the aircraft through a previously approved electrical connection. If modification to the ship's system is necessary to support this installation, additional minor modifications with appropriate approval is necessary.

1.4. Installation Introduction

This manual presents the installation instructions for the Airfilm Camera Systems model AFNW-1 Utility-mount for the Eurocopter AS350 AND AS355 series of Rotorcraft. The mount is designed to facilitate the attachment of equipment such as FLIR cameras, video cameras, searchlights, etc.

1.5 General

These instructions cover the AirFilm Nose Window Mount installation on the Eurocopter AS350 and AS355 series rotorcraft.

Precautions:
All precautions will be in **bold face**

Referenced publications
AC 43.13-2 and AC43.13-1B

Referenced drawing
AFNW-1

Distribution:
Installation instructions shall accompany the maintenance manuals of aircraft on which the mount is installed.

Definitions / Abbreviations:
IAW: in accordance with

Standards of measurement:
All measurements in 100ths of an inch
All weights in US pounds
All torques in inch pounds

1.6 Control & Operational Information

Special procedures / precautions:

Maximum mount payload not to exceed 200 lbs (90 kg), or 3.6 square feet (3345 square cm) projected area.

2.0 Installation Information

- Installation of mount assembly:
- Reference drawing AFNW-1

The following drawing shows the AFNW-1 Mount. The AFNW-1 Mount consists of a welded steel structure that is attached to the floor airframe structure with quantity 15, AN 3 fasteners. The front of the mount protrudes through the modified plexiglas chin bubble and the Gyron Camera is attached to the flat plate portion of the mount.

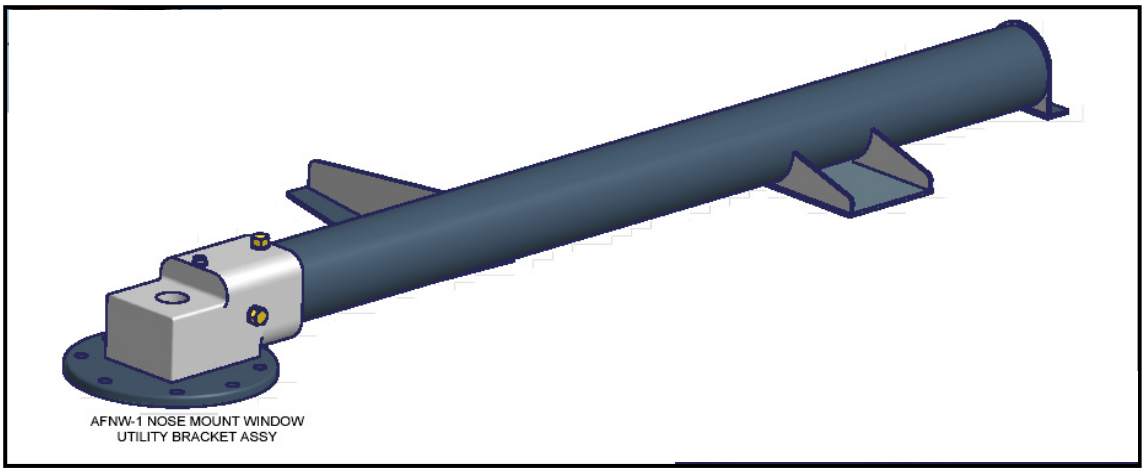

Figure 1. AFNW-1 NOSE WINDOW MOUNT

Figure 2 LEFT SIDE FLOOR OF HELICOPTER

3.0 Installation Procedures

3.0.1 Remove blank bolts.
3.0.2 Remove left chin bubble per Helicopter Maintenance Manual instructions.
3.0.3 Locate 3 rivets in floor in front of seat belt hard point per the following figure.

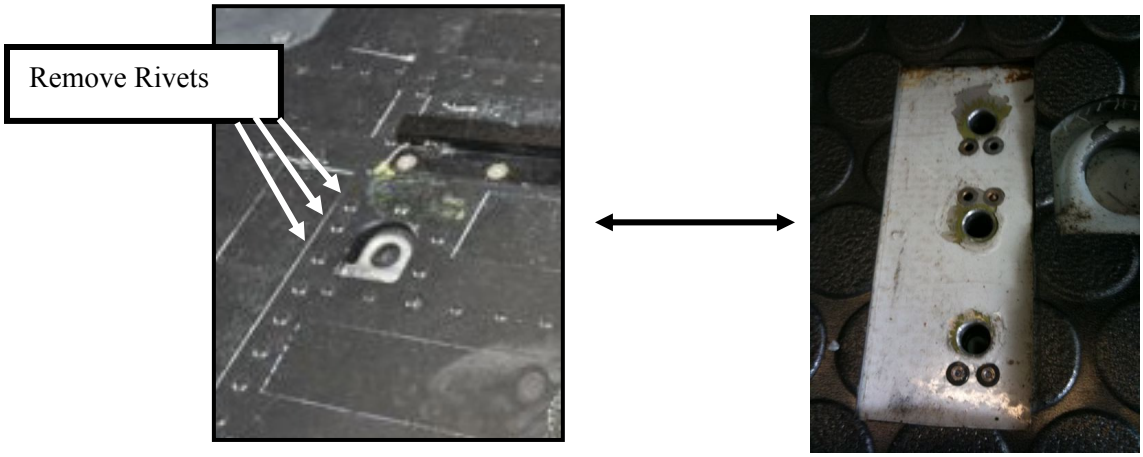

Figure 3. Front of seat hard pan

3.0.4 Drill out button head rivets and install 3 ea , NAS1766L3 nut plates (Corner floating type) or equivalent.
3.0.5 Position mount on floor with aft bracket located for adequate edge distance for AN3 Bolt (.30 inches).
3.0.6 Outline mount on the floor.
3.0.7 Mark out button head rivets on the forward and middle arms to be removed for flat surface for mount arm location and. nut plate installation.

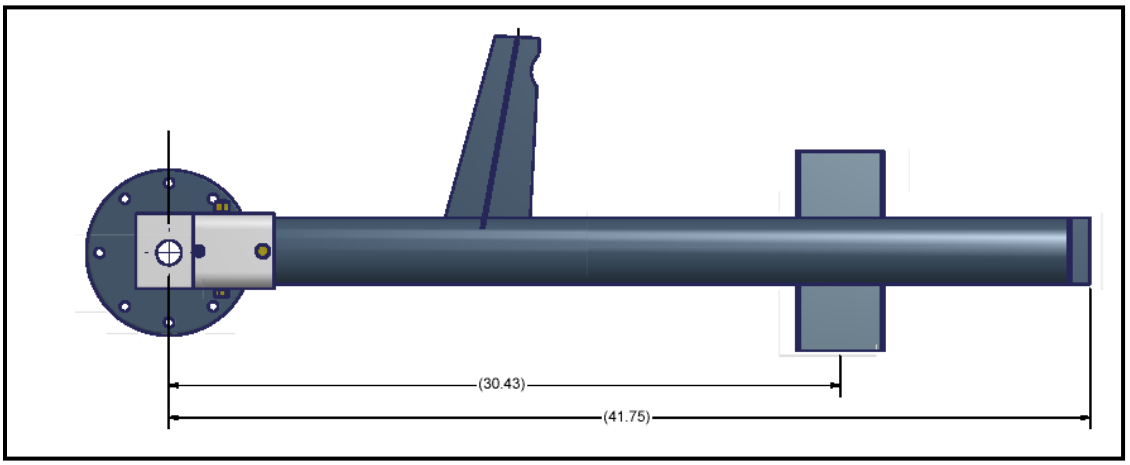

Figure 4. Window Nose Mount

3.0.8 Drill out marked rivets.
3.0.9 Install flush NAS 1097 or MS20427 rivets (dia. and length as required) in all holes under the mount (Except the holes for the nut plates).

Note: De-Bur all holes, treat all bare aluminum with Alodine per Mil-DTL-5541 as required

3.0.10 Re position mount on floor in final position.
3.0.11 Mark holes from bottom of the floor onto the mount.
3.0.12 Check hole locations on mount for adequate edge distance of 1.2 to 2.0 time AN3 bolt dia.
3.0.13 Drill mount at marked location using rivet hole diameter.
3.0.14 Re position mount on floor in final position and cleco in place.
3.0.15 Line drill the mount and floor with #10 drill.
3.0.16 Redrill holes in floor with #8 drill

Note: De-Bur all holes, treat all bare aluminum with Alodine per Mil-DTL-5541 as required

3.0.17 Install 12 ea MS21059L3 or NAS1766L3 (Corner floating type) or equivalent in middle and forward arm locations
3.0.18 Install Mount using quantity 15, AN3 Bolts, AN960-10 or 10L washers
3.0.19 Install Modified nose window per Helicopter Maintenance Manual

3.0.20 Installation Complete

The adjacent photograph shows the mount installed on the floor of the helicopter

Figure 5. CoPilot's Seat Left Side installation

4.0 Removal Procedures

4.0.1 Remove 15 each, AN3 Bolts and AN960-10 or -10L washers

4.0.2 Remove Mount

4.0.3 Remove modified nose window.

4.0.4 Install original chin bubble and blank bolts.

4.0.6 Reinstall quantity 15, AN3 bolts into nut plates

4.0.7 Removal complete

The following photograph shows helicopter floor with bolts installed in existing nut plates

Figure 6. Copilot's left side view

5.0 Weight and Balance

Table 5.1: AFNW-1 Mount Weight and Center of Gravity

The following is the installed weight and center of gravity for the AFNW-1 Utility Mount and the payload location.

PART NO.	DESCRIPTION	WEIGHT	STATION (inches) (mm	BL (inches) (mm
AFNW-1	Nose Window Mount	25 lbs 11 kg	20 inches 508 mm	-22 inches (left) -559 mm (left)

Table 5.2: Payload Size, Weight and Center of Gravity Limit

The following is the installed weight and center of gravity for the AFNW-1 Utility Mount and the payload location.

DESCRIPTION	PROJECTED AREA	WEIGHT	STATION (inches) (mm	BL (inches) (mm
Maximum Weight	3.6 sq. feet 3345 cm^2	200 lbs 90 kg	5 inches 127 mm	-22 in (left) -559 mm(left)

APPENDIX A - METHOD OF ADDING ADDITIONAL SENSOR /CAMERA / PAYLOADS

1. Overview

This Appendix provides the requirements necessary to qualify additional sensor / camera / light payloads not listed in the front of this manual. It may also be used as a check list for previously approved sensor /cameras / light payloads if desired.

The STC flight testing was conducted and the STC approved with the largest and heaviest payload expected for use with this mount. The specific sensor/cameras/light not listed in the installation manual of equal or lesser than the limit case are accepted with this follow-on test plan.

2. Sensor/ Camera/ payload

Make & Model _____

3. Test Team

Pilot/s

_____ _____
Print Name

Mechanic and/or Engineer and/or Camera Operator

_____ _____
Print Name

4. Test Aircraft Configuration and Location

Aircraft Model, Registration & Serial Number

_____ _____ _____
Model Registration Number Serial Number

Test Configurations
Empty weight with appropriate fuel and camera system installed
Takeoff Gross weight with crew

Configuration	Gross Weight	Longitudinal CG	Lateral CG
Empty Wt			
Takeoff Wt			

Test Location

_____
Airport or Test Site

5. Test Conditions

Date: _____

Weather: Ceiling _____ Visibility _____ Winds _____

Altimeter _____ Field Elevation _____

Flight Time: Engine Start _____ Shut Down _____ Flt Time _____

6. Flight Test

6.1 Overview

Applicable regulations demonstrated for compliance are indicated with the following symbol ➔. The testing required for the compliance findings of this installation will be made by as a subject/qualitative evaluation. Although the most critical CG is considered to be at the aft limit for most tests this configuration is mounted forward of the mast should not approach the aft limits. This also depends on crew loading. The test team conducts the following tests and evaluations and mark initial the box at the end of each section if the configuration successfully passes the requirements.

6.2 FAR § 27.51 Takeoff

6.2.1 Applicable Regulation

➔(a) The takeoff, with takeoff power and rpm, and with the extreme forward center of gravity -
➔(1) May not require exceptional piloting skill or exceptionally favorable conditions; and
(2) Must be made in such a manner that a landing can be made safely at any point along the flight path if an engine fails.
(b) Paragraph (a) of this section must be met throughout the ranges of -
(1) Altitude, from standard sea level conditions to the maximum altitude capability of the rotorcraft, or 7,000 feet, whichever is less; and
(2) Weight, from the maximum weight (at sea level) to each lesser weight selected by the applicant for each altitude covered by paragraph (b)(1) of this section.

6.2.2 Method of Compliance

The recommended takeoff procedure must be demonstrated to remain clear of the HV "avoid" areas without requiring exceptional piloting skill or exceptionally favorable conditions.

A qualitative evaluation of the ability to safely land at any point along the flight path will be made using judgment and experience with the basic aircraft. No engine failure testing at low altitude will be conducted.

The normal takeoff procedures will be used for the sensor/camera/light payload and mount installation.

6.2.3 Findings

Satisfactory ☐

6.3 FAR § 27.71 Glide Performance

6.3.1 Applicable Regulation

➔For single engine helicopters and multiengine helicopters that do not meet the Category A engine isolation requirements of Part 29 of this chapter, the minimum rate of descent airspeed and the best angle of glide airspeed must be determined in autorotation at -
(a) Maximum weight; and
(b) Rotor speed(s) selected by the applicant.

6.3.2 Method of Compliance

(1) Performance capabilities during stabilized autorotative descent are useful tools to assist the pilot when all engines fail. This information is also useful in determining the suitability of available landing areas along a given route segment.

(2) Two speeds are of particular importance, the speed for minimum rate of descent and the speed for best angle of glide. These speeds along with glide distance information are required as flight manual entries per FAR § 27.1587.

The best angle of glide performance will be evaluated at a single speed and low power (needles joined) descent. An autorotative descent starting at least 1000 feet above the ground and at the speed published in the RFM, 100% RPM value will be demonstrated. Small turns will be conducted in the descent.

The aircraft should be easily controllable and the difference between the mount and camera/sensor/light payload and the clean configuration is the evaluation point.

6.3.3 Findings

Satisfactory ☐ Altitude Band H_P _____ Fuel Gage Reading _____

6.4 FAR § 27.143 Controllability and Maneuverability

6.4.1 Applicable Regulation

➔(a) The rotorcraft must be safely controllable and maneuverable -
- ➔ (1) During steady flight; and
- ➔(2) During any maneuver appropriate to the type, including -
  - ➔ (i) Takeoff;
  - ➔ (ii) Climb;
  - ➔ (iii) Level flight;
  - ➔ (iv) Turning flight;
  - (v) Glide;
  - ➔ (vi) Landing (power on and power off); and
  - (vii) Recovery to power on flight from a balked autorotative approach.

➔(b) The margin of cyclic control must allow satisfactory roll and pitch control at VNE with -
- (1) Critical weight;
- (2) Critical center of gravity;
- (3) Critical rotor rpm; and
- (4) Power off (except for helicopters demonstrating compliance with paragraph (e) of this section) and power on.

(c) A wind velocity of not less than 17 knots must be established in which the rotorcraft can be operated without loss of control on or near the ground in any maneuver appropriate to the type (such as crosswind takeoffs, sideward flight, and rearward flight), with -
- (1) Critical weight;
- (2) Critical center of gravity;
- (3) Critical rotor rpm; and
- (4) Altitude, from standard sea level conditions to the maximum altitude capability of the rotorcraft or 7,000 feet, whichever is less.

(d) The rotorcraft, after failure of one engine in the case of multiengine rotorcraft that meet Transport Category A engine isolation requirements, or complete engine failure in the case of other rotorcraft, must be controllable over the range of speeds and altitudes for which certification is

requested when such power failure occurs with maximum continuous power and critical weight. No corrective action time delay for any condition following power failure may be less than -

(1) For the cruise condition, one second, or normal pilot reaction time (whichever is greater); and

(2) For any other condition, normal pilot reaction time.

(e) For helicopters for which a VNE (power off) is established under § 27.1505(c), compliance must be demonstrated with the following requirements with critical weight, critical center of gravity, and critical rotor rpm:

(1) The helicopter must be safely slowed to VNE (power off), without exceptional pilot skill, after the last operating engine is made inoperative at power on VNE.

(2) At a speed of 1.1 VNE (power off), the margin of cyclic control must allow satisfactory roll and pitch control with power off.

6.4.2 Method of Compliance

The general requirements for control and for maneuverability are summarized in section (a), which is largely self-explanatory.

Section (b) specifies flight at V_{NE} with critical weight, center of gravity (CG), rotor RPM, and power. Adequate cyclic authority must remain at V_{NE} for nose down pitching of the rotorcraft and for adequate roll control.

The helicopter will be flown between 1000 and 3000 feet above ground. The test altitude will be dependent on traffic and terrain and conditions close to sea level pressure are desirable. V_{NE} will be the value stated in the RFM for the test density altitude.

Qualitative measurement techniques (pilot opinion) will be used. The tests will include:

Takeoff
Climbing flight
Forward flight to V_{NE} at MCP (maybe less than MCP)
Left & right 30 degree bank turns at V_{NE} and at MCP (maybe less than MCP)
Take-off & Landings (Power on only).

The aircraft should be easily controllable and adequate cyclic margins should exist throughout the flight test points. The difference between the mount and sensor / camera / light payload and the clean configuration is the evaluation point.

6.4.3 Findings

Satisfactory ☐ Cruise Altitude H_P _____ Fuel Gage Reading _____

6.5 FAR § 27.171 Stability: General

6.5.1 Applicable Regulation

➔The rotorcraft must be able to be flown, without undue pilot fatigue or strain, in any normal maneuver for a period of time as long as that expected in normal operation. At least three landings and takeoffs must be made during this demonstration.

6.5.2 Method of Compliance

Compliance with the requirements of this section can often be obtained for the VFR condition without any specific or designated flight testing. This test should be conducted with minimum required systems in the aircraft and with minimum flight crew.

Compliance for this requirement will be evaluated throughout the test program.

6.5.3 Findings

Satisfactory ☐

6.6 FAR § 27.251 Vibration

6.6.1 Applicable Regulation

➔Each part of the rotorcraft must be free from excessive vibration under each appropriate speed and power condition.

6.6.2 Method of Compliance

This flight requirement may be both a qualitative and quantitative flight evaluation. Section 27.571(a) contains the flight load survey requirement that results in accumulation of vibration quantitative data. Section 27.629 generally requires quantitative data to show freedom from flutter for each part of the rotorcraft including control or stabilizing surfaces and rotors.

The aircraft should have a good track & balance for this evaluation. The airspeed should evaluated at 20 kt increments out to the RFM V_{NE} speed. Variations in rotor RPM expected in normal flight should be evaluated. Changes in vibration are best sensed in the cyclic and pedal controls. The stability of the camera/sensor image will be a good indicator.

The pilot will make a subjective evaluation of the difference between the mount and sensor / camera/ light payload and the clean configuration is the evaluation point.

Compliance with this requirement will be evaluated during testing of FAR § 27.143 Controllability and Maneuverability.

6.6.3 Findings

Satisfactory ☐

6.7 FAR § 27.773 Pilot Compartment View

6.7.1 Applicable Regulation

(a) Each pilot compartment must be free from glare and reflections that could interfere with the pilot's view, and designed so that--
➔ (1) Each pilot's view is sufficiently extensive, clear, and undistorted for safe operation; and
(2) Each pilot is protected from the elements so that moderate rain conditions do not unduly impair his view of the flight path in normal flight and while landing.
➔ (b) If certification for night operation is requested, compliance with paragraph (a) of this section must be shown in night flight tests.

6.7.2 Method of Compliance

The section outlines requirements for pilot view in fairly general terms. The aircraft was approved with the installed glareshield and instrument panel that meet the rules. Any additional equipment/monitors must be positioned so as not to limit or obstruct the pilot's field of view. There will be some cases where the installation will be temporary and for a unique mission and consideration should be given for these limited cases and time.

If night operations are expected with an operational system, a "dark cockpit" or night evaluation will be necessary to insure the glare/reflection will not interfere with the pilot duties. A limitation to the use at night is an option.

6.7.3 Findings

Satisfactory ☐

6.8 FAR § 27.787 Cargo & Baggage Compartment

6.8.1 Applicable Regulation

Cargo and baggage compartments.

(a) Each cargo and baggage compartment must be designed for its placarded maximum weight of contents and for the critical load distributions at the appropriate maximum load factors corresponding to the specified flight and ground load conditions, except the emergency landing conditions of Sec. 27.561.

(b) There must be means to prevent the contents of any compartment from becoming a hazard by shifting under the loads specified in paragraph (a) of this section.

➔ **[**(c) Under the emergency landing conditions of Sec. 27.561, cargo and baggage compartments must--

(1) Be positioned so that if the contents break loose they are unlikely to cause injury to the occupants or restrict any of the escape facilities provided for use after an emergency landing; or

(2) Have sufficient strength to withstand the conditions specified in Sec. 27.561 including the means of restraint, and their attachments, required for the maximum authorized weight of cargo and baggage at the critical loading distribution.**]**

(d) If cargo compartment lamps are installed, each lamp must be installed so as to prevent contact between lamp bulb and cargo.

6.8.2 Method of Compliance

Amendment 27-27 adds two subparagraphs to § 27.787(c) which clarify that cargo and baggage compartments should be designed to protect occupants from injury by the compartment contents during emergency landings. This may be done by location or by retention provisions.

The sensor/camera/light controllers and power supply must be located and secured in a position that will not endanger occupants in an emergency landing impact.

Consideration should be given to stowage and egress when filming in hovering flight. In some cases this might not be possible.

6.8.3 Findings

Comment: _____

Satisfactory ☐

6.9 FAR § 27.1301 Function and Installation.

6.9.1 Applicable Regulation

Each item of installed equipment must--
➔ (a) Be of a kind and design appropriate to its intended function;
(b) Be labeled as to its identification, function, or operating limitations, or any applicable combination of these factors;
(c) Be installed according to limitations specified for that equipment; and
➔ (d) Function properly when installed.

6.9.2 Method of Compliance

For optional equipment, the emphasis on functioning is rather limited compared to that for required equipment. The conditions under which the optional equipment is evaluated should be recorded in the report. The major emphasis for this type of equipment should be to ensure it does not interfere with the

operation of systems that are required for safe operation of the rotorcraft, and that the failure modes are acceptable and do not create any hazards.

During flight operations, operate all avionics and electrical systems. Complete the matrix below. The matrix is laid out with the newly installed equipment listed at the top of the page and all aircraft systems listed down the left side of the page. Note any EMI or RFI either TO or FROM the installed equipment. Note any anomalies or EMI/RFI interference to other instruments or indications during all testing phases of flight.

Each item must be checked. Check off each block if no interference is noted. If interference is present during the test, DO NOT CHECK THE BOX and explain in Comments section at end of section. If applicable, note relevant conditions (i.e. frequencies, OBI selection, function modes) under which the interference occurred.

6.9.3 Findings

Interference? 	**Camera/Sensor/Light**	**Position Controller**
Camera/Sensor/Light		
Position Controller		
VHF Comm 1		
VHF Comm 2		
VHF Comm 3		
VHF NAV 1		
VHF NAV 2		
ADF 1		
XPONDER 1		
Other Radios		
Audio 1		
Audio 2		
Standby Compass		
Engine Inst		
Fuel Gage		
Clock		
Voltmeter		
Ammeter		
Other		

EMI / RFI Comments:

Satisfactory ☐

Signatures

General test findings _____

Pilot Signature _____

Mechanic/ Engineer _____

Other Flt Personnel Signature & Function

_____